

### B. Availability of Rulemaking Documents

An electronic copy of rulemaking documents may be obtained from the internet by—

1. Searching the Federal eRulemaking Portal (<http://www.regulations.gov>);
2. Visiting the FAA's Regulations and Policies web page at [http://www.faa.gov/regulations\\_policies](http://www.faa.gov/regulations_policies) or
3. Accessing the Government Printing Office's web page at <http://www.gpo.gov/fdsys/>.

Copies may also be obtained by sending a request to the Federal Aviation Administration, Office of Rulemaking, ARM-1, 800 Independence Avenue SW, Washington, DC 20591, or by calling 202-267-9677. Commenters must identify the docket or notice number of this rulemaking.

All documents the FAA considered in developing this proposed rule, including economic analyses and technical reports, may be accessed from the internet through the Federal eRulemaking Portal referenced in item (1) above.

### List of Subjects in 14 CFR Part 25

Aircraft, Aviation safety, Reporting and recordkeeping requirements.

### The Proposed Amendment

In consideration of the foregoing, the Federal Aviation Administration proposes to amend chapter I of title 14, Code of Federal Regulations as follows:

### PART 25—AIRWORTHINESS STANDARDS: TRANSPORT CATEGORY AIRPLANES

- 1. The authority citation for part 25 continues to read as follows:

**Authority:** 49 U.S.C. 106(f), 106(g), 40113, 44701, 44702 and 44704.

- 2. Amend § 25.841 by revising paragraphs (a) introductory text and (b)(6) and adding paragraphs (c) and (d) to read as follows:

#### § 25.841 Pressurized cabins.

(a) Except as provided in paragraph (c) of this section, pressurized cabins and compartments to be occupied must be equipped to provide a cabin pressure altitude of not more than 8,000 feet under normal operating conditions.

\* \* \* \* \*

(b) \* \* \*

(6) Warning indication at the pilot or flight engineer station to indicate when the safe or preset pressure differential and cabin pressure altitude limits are exceeded. Appropriate warning markings on the cabin pressure differential indicator meet the warning

requirement for pressure differential limits, and an alert meets the warning requirement for cabin pressure altitude limits, if it warns the flightcrew when the cabin pressure altitude exceeds 10,000 feet, except as provided in paragraph (d) of this section.

\* \* \* \* \*

(c) When operating into or out of airports with elevations at or above 8,000 feet, the cabin pressure in pressurized cabins and occupied compartments may be equal to or less than the airport elevation provided:

(1) The airplane is being operated at or below 25,000 feet; and

(2) The cabin pressurization system is designed to minimize the time in flight that passenger cabin occupants may be exposed to cabin pressure altitudes exceeding 8,000 feet.

(d) When operating into or out of airports with elevations exceeding 8,000 feet and the airplane is at or below 25,000 feet, the cabin altitude warning alert may be provided at 15,000 feet, or 2,000 feet above the elevation, whichever is greater, provided that:

(1) An alert is provided to clearly indicate to the flightcrew that the cabin high altitude warning has shifted above 10,000 feet;

(2) If the cabin altitude warning alert is shifted above 10,000 feet automatically, an alert is provided to notify the flightcrew to take action should the automatic shift function fail; and

(3) Either an alerting system is installed to notify the flightcrew members on flight deck duty when to don oxygen in accordance with the applicable operating regulations; or flight procedures acceptable to the FAA administrator are provided in the airplane flight manual that require the pilot flying to don oxygen when the high altitude cabin warning has shifted above 10,000 feet and require other flightcrew members on flight deck duty to monitor the cabin pressure to utilize oxygen in accordance with the applicable operating regulations.

- 3. Amend § 25.1447 by revising paragraph (c)(1) and adding paragraph (c)(5) to read as follows:

#### § 25.1447 Equipment standards for oxygen dispensing units.

\* \* \* \* \*

(c) \* \* \*

(1) There must be an oxygen dispensing unit connected to oxygen supply terminals immediately available to each occupant wherever seated, and at least two oxygen-dispensing units connected to oxygen terminals in each lavatory. The total number of dispensing units and outlets in the cabin must

exceed the number of seats by at least 10 percent. The extra units must be as uniformly distributed throughout the cabin as practicable. Except as provided in paragraph (c)(5) of this section, if certification for operation above 30,000 feet is requested, the dispensing units providing the required oxygen flow must be automatically presented to the occupants before the cabin pressure altitude exceeds 15,000 feet. The crewmembers must be provided with a manual means of making the dispensing units immediately available in the event of failure of the automatic system.

\* \* \* \* \*

(5) When operating into or out of airports with elevations at or above 8,000 feet, the dispensing units providing the required oxygen flow may be automatically presented to the occupants at 15,000 feet or within 2,000 feet of the airport elevation, whichever is higher, provided the airplane is being operated at altitudes at or below 25,000 feet.

Issued under authority provided by 49 U.S.C. 106(f) and 44701(a) in Washington, DC, on March 29, 2019.

**Earl Lawrence,**  
Executive Director, Aircraft Certification Service.

[FR Doc. 2019-06765 Filed 4-4-19; 8:45 am]

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## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA-2019-0240; Product Identifier 2018-CE-057-AD]

RIN 2120-AA64

### Airworthiness Directives; Pilatus Aircraft Ltd. Airplanes

**AGENCY:** Federal Aviation Administration (FAA), Department of Transportation (DOT).

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** We propose to adopt a new airworthiness directive (AD) for certain Pilatus Aircraft Ltd. Models PC-6, PC-6/350, PC-6/350-H1, PC-6/350-H2, PC-6/A, PC-6/A-H1, PC-6/A-H2, PC-6/B-H2, PC-6/B1-H2, PC-6/B2-H2, PC-6/B2-H4, PC-6/C-H2, PC-6/C1-H2, PC-6-H1, and PC-6-H2 airplanes. This proposed AD results from mandatory continuing airworthiness information (MCAI) originated by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes

the unsafe condition as flap actuator taper pins that were not swaged during the manufacturing process. We are issuing this proposed AD to require actions to address the unsafe condition on these products.

**DATES:** We must receive comments on this proposed AD by May 20, 2019.

**ADDRESSES:** You may send comments by any of the following methods:

- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.
- *Fax:* (202) 493-2251.
- *Mail:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE, Washington, DC 20590.
- *Hand Delivery:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE, Washington, DC 20590, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this proposed AD, contact PILATUS Aircraft Ltd., Customer Technical Support (MCC), P.O. Box 992, CH-6371 Stans, Switzerland; phone: +41 (0)41 619 67 74; fax: +41 (0)41 619 67 73; email: [techsupport@pilatus-aircraft.com](mailto:techsupport@pilatus-aircraft.com); internet: <http://www.pilatus-aircraft.com>. You may review this referenced service information at the FAA, Policy and Innovation Division, 901 Locust, Kansas City, Missouri 64106. For information on the availability of this material at the FAA, call (816) 329-4148.

#### Examining the AD Docket

You may examine the AD docket on the internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2018-0240; or in person at Docket Operations between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for Docket Operations (telephone (800) 647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

#### FOR FURTHER INFORMATION CONTACT:

Doug Rudolph, Aerospace Engineer, FAA, Small Airplane Standards Branch, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329-4059; fax: (816) 329-4090; email: [doug.rudolph@faa.gov](mailto:doug.rudolph@faa.gov).

#### SUPPLEMENTARY INFORMATION:

#### Comments Invited

We invite you to send any written relevant data, views, or arguments about

this proposed AD. Send your comments to an address listed under the **ADDRESSES** section. Include “Docket No. FAA-2018-0240; Product Identifier 2018-CE-057-AD” at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD because of those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

#### Discussion

The European Aviation Safety Agency (EASA), which is the Technical Agent for the Member States of the European Community, has issued EASA AD No. 2018-0235, dated November 5, 2018 (referred to after this as “the MCAI”), to correct an unsafe condition for Pilatus Aircraft Limited Models PC-6, PC-6/350, PC-6/350-H1, PC-6/350-H2, PC-6/A, PC-6/A-H1, PC-6/A-H2, PC-6/B-H2, PC-6/B1-H2, PC-6/B2-H2, PC-6/B2-H4, PC-6/C-H2, PC-6/C1-H2, PC-6-H1, and PC-6-H2 airplanes. The MCAI states:

During a recent overhaul, two new flap actuators were found to have taper pins installed that, apparently, had not been swaged. Investigation results identified that the taper pins had been incorrectly swaged during the manufacturing process.

This condition, if not detected and corrected, could lead to loss of one or both taper pins, consequent asymmetric flap deployment or flap surface flutter, possibly resulting in loss of control of the aeroplane.

To address this potential unsafe condition, Pilatus issued the [service bulletin] SB to provide inspection instructions.

For the reason described above, this [EASA] AD requires a one-time inspection of the taper pins of the affected parts for correct installation and, depending on findings, accomplishment of applicable corrective action(s). This [EASA] AD also requires inspection of, and, depending on findings, corrective action(s) on, affected parts held as spare, prior to installation.

You may examine the MCAI on the internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2018-0240.

#### Related Service Information Under 1 CFR Part 51

Pilatus Aircraft Ltd. has issued Pilatus PC-6 Service Bulletin No. 27-005, dated July 2, 2018. The service information contains procedures for removing and

inspecting the flap actuator assemblies and pushrod assemblies, modifying or replacing the taper pins if necessary, and reinstalling the assemblies. This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the **ADDRESSES** section.

#### FAA's Determination and Requirements of the Proposed AD

This product has been approved by the aviation authority of another country, and is approved for operation in the United States. Pursuant to our bilateral agreement with this State of Design Authority, they have notified us of the unsafe condition described in the MCAI and service information referenced above. We are proposing this AD because we evaluated all information and determined the unsafe condition exists and is likely to exist or develop on other products of the same type design.

#### Costs of Compliance

We estimate that this proposed AD will affect 30 products of U.S. registry. We also estimate that it would take about 12 work-hours per product to comply with the basic inspection requirements of this proposed AD. The average labor rate is \$85 per work-hour.

Based on these figures, we estimate the cost of the proposed AD on U.S. operators to be \$30,600, or \$1,020 per product.

In addition, we estimate that any necessary follow-on modification or replacement actions would require parts costing \$30,000, for a cost of \$1,000 per product. We have no way of determining the number of products that may need these actions.

#### Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. “Subtitle VII: Aviation Programs,” describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in “Subtitle VII, Part A, Subpart III, Section 44701: General requirements.” Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on

products identified in this rulemaking action.

This AD is issued in accordance with authority delegated by the Executive Director, Aircraft Certification Service, as authorized by FAA Order 8000.51C. In accordance with that order, issuance of ADs is normally a function of the Compliance and Airworthiness Division, but during this transition period, the Executive Director has delegated the authority to issue ADs applicable to small airplanes, gliders, balloons, airships, domestic business jet transport airplanes, and associated appliances to the Director of the Policy and Innovation Division.

### Regulatory Findings

We determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this proposed regulation:

- (1) Is not a “significant regulatory action” under Executive Order 12866,
- (2) Is not a “significant rule” under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),
- (3) Will not affect intrastate aviation in Alaska, and
- (4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

### The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

### PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

- 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

**Pilatus Aircraft Ltd.:** Docket No. FAA–2018–0240; Product Identifier 2018–CE–057–AD.

#### (a) Comments Due Date

We must receive comments by May 20, 2019.

#### (b) Affected ADs

None.

#### (c) Applicability

This AD applies to Pilatus Aircraft Ltd. Models PC–6, PC–6/350, PC–6/350–H1, PC–6/350–H2, PC–6/A, PC–6/A–H1, PC–6/A–H2, PC–6/B–H2, PC–6/B1–H2, PC–6/B2–H2, PC–6/B2–H4, PC–6/C–H2, PC–6/C1–H2, PC–6–H1, and PC–6–H2 airplanes, all serial numbers, certificated in any category, with a left-hand or right-hand flap actuator assembly part number (P/N) 6132.0039.51 or P/N 6132.0039.52 or pushrod assembly P/N 6132.0040.00 installed, except those assemblies supplied by Pilatus Aircraft Ltd. with a European Aviation Safety Agency (EASA) form 1 tag dated July 2, 2018 or later.

**Note 1 to paragraph (c) of this AD:** These airplanes may also be identified as Fairchild Republic Company airplanes, Fairchild Industries airplanes, Fairchild Heli Porter airplanes, or Fairchild-Hiller Corporation airplanes.

#### (d) Subject

Air Transport Association of America (ATA) Code 27: Flight Controls.

#### (e) Reason

This AD was prompted by mandatory continuing airworthiness information (MCAI) originated by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as flap actuator taper pins that were not swaged during the manufacturing process. We are issuing this AD to prevent loss of one or both taper pins that could lead to asymmetric flap deployment or flap surface flutter and result in loss of control of the airplane.

#### (f) Actions and Compliance

Unless already done, do the following actions in paragraphs (f)(1) and (2) of this AD:

- (1) Within the next 100 hours time-in-service after the effective date of this AD or within the next 12 months after the effective date of this AD, whichever occurs first, prepare the airplane and inspect each flap actuator taper pin for correct installation by following the Accomplishment Instructions-Part 1-On Aircraft, paragraphs 3.A through 3.B(2), of Pilatus Aircraft Ltd. PC–6 Service Bulletin No. 27–005, dated July 2, 2018 (Pilatus SB No. 27–005).

- (i) If a taper pin has any damage, before further flight, replace and swage the taper pin and reinstall the pushrod assembly by following the Accomplishment Instructions-Part 1-On Aircraft, paragraphs 3.C and 3.D of Pilatus SB No. 27–005.
  - (ii) If a taper pin is incorrectly swaged or is not swaged, before further flight, swage the

taper pin and reinstall the pushrod assembly by following the Accomplishment Instructions-Part 1-On Aircraft, paragraphs 3.C and 3.D of Pilatus SB No. 27–005.

- (2) After the effective date of this AD, do not install a flap actuator assembly, P/N 6132.0039.51 or P/N 6132.0039.52, or pushrod assembly P/N 6132.0040.00 on any airplane unless the part was supplied by Pilatus Aircraft Ltd. with an EASA form 1 tag dated July 2, 2018 or later, or the part has been inspected in accordance with paragraphs (f)(1)(i) and (ii) of this AD.

#### (g) Other FAA AD Provisions

The following provisions also apply to this AD:

- (1) *Alternative Methods of Compliance (AMOCs):* The Manager, Small Airplane Standards Branch, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. Send information to ATTN: Doug Rudolph, Aerospace Engineer, FAA, Small Airplane Standards Branch, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329–4059; fax: (816) 329–4090; email: [doug.rudolph@faa.gov](mailto:doug.rudolph@faa.gov). Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

- (2) *Contacting the Manufacturer:* For any requirement in this AD to obtain corrective actions from a manufacturer, the action must instead be accomplished using a method approved by the Manager, Small Airplane Standards Branch, FAA, or EASA.

#### (h) Related Information

Refer to MCAI EASA AD No. 2018–0235, dated November 5, 2018, for related information. You may examine the MCAI on the internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA–2018–0240. For service information related to this AD, contact PILATUS Aircraft Ltd., Customer Technical Support (MCC), P.O. Box 992, CH–6371 Stans, Switzerland; phone: +41 (0)41 619 67 74; fax: +41 (0)41 619 67 73; email: [techsupport@pilatus-aircraft.com](mailto:techsupport@pilatus-aircraft.com); internet: <http://www.pilatus-aircraft.com>. You may review this referenced service information at the FAA, Policy and Innovation Division, 901 Locust, Kansas City, Missouri 64106. For information on the availability of this material at the FAA, call (816) 329–4148.

Issued in Kansas City, Missouri, on March 25, 2019.

**Melvin J. Johnson,**

*Aircraft Certification Service, Deputy Director, Policy and Innovation Division, AIR–601.*

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